

RECOMMENDED FLIGHT MANUAL

COVER PAGE

Nationality and Identification Number: _____

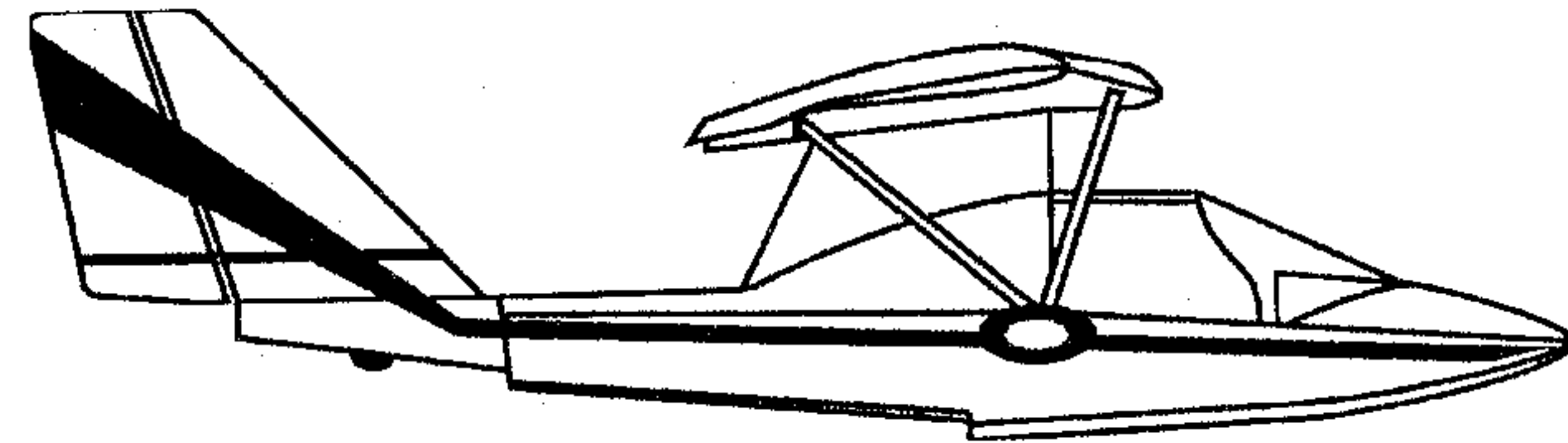
Manufacturer/Builder: _____

Address of Manufacturer/Builder: _____

Aircraft Type: _____

Kit Manufacturer: Progressive Aerodyne, Inc.
520 Clifton St.
Orlando, FL 32808
U.S.A.

Kit Manufacturer's Serial Number: _____



GENERAL AMENDMENT RECORD SHEET

AMENDMENT NUMBER	PARAGRAPH(S) AFFECTED	SIGNATURE	DATE OF INCORPORATION

NOTE: Amendment numbers may not always be consecutive.

INTRODUCTION

This Flight Manual contains material to be supplied by the owner/builder ("Test Aircraft"), as well as supplemental data supplied by Progressive Aerodyne, Inc. Sample data from the Progressive Aerodyne Demonstration SeaRey ("P/A Demo") is included for reference.

This Flight Manual applies only to the particular aircraft identified by the Identification Number and Serial Number on the cover page and contains limitations and essential data for this aircraft.

This aircraft is registered as an experimental-type aircraft on the basis of the equipment fitted at the time of issue of the Airworthiness Certificate. Any changes in equipment must be listed in the aircraft's logbook and appropriate amendments should be made in this Manual.

FLIGHT MANUAL – TABLE OF CONTENTS

TABLE OF CONTENTS	SECTION
AIRCRAFT GENERAL DATA	1
LIMITATIONS	2
EMERGENCY PROCEDURES	3
NORMAL PROCEDURES	4
PERFORMANCE	5
WEIGHT & BALANCE (LOADING DATA)	6
AIRCRAFT & SYSTEMS DESCRIPTIONS	7
AIRCRAFT HANDLING, SERVICE & MAINTENANCE	8
SUPPLEMENTS	9

SECTION 1 – AIRCRAFT GENERAL DATA

TABLE OF CONTENTS	PAGE
ENGINE	1.1
PROPELLER	1.1
FUEL	1.1
OIL	1.1
MAXIMUM APPROVED WEIGHTS	1.2
BASIC EMPTY WEIGHT	1.2
ABBREVIATIONS, DEFINITIONS & TERMINOLOGY	1.2
WEIGHT & BALANCE TERMINOLOGY	1.3

SECTION 1 – AIRCRAFT GENERAL DATA, CONTINUED

MAXIMUM APPROVED WEIGHTS

	Test Aircraft	(P/A Demo)
Gross Weight:	_____	1370 lbs
Max Take-off Weight:	_____	1370 lbs
Max Landing Weight:	_____	1370 lbs
Max Baggage Weight:	_____	55 lbs
Basic Empty Weight:	_____	850 lbs

ABBREVIATIONS & DEFINITIONS

The following shall apply throughout this Manual:

PRESSURE ALTITUDE is the altitude read from an altimeter when the altimeter's barometric scale has been set to 29.92 H.g. (inches of mercury).

IAS (Indicated Airspeed) is the speed shown on the airspeed indicator.

TAKE-OFF SAFETY SPEED is a speed chosen to ensure that adequate control will exist under all conditions, including turbulence and sudden and complete engine failure, during the climb after take-off.

APPROACH SPEED is a speed chosen to ensure that adequate control exists under all conditions, including turbulence, to carry out a normal flare and touchdown.

V_c – NORMAL OPERATING LIMIT SPEED is the speed that shall normally not be exceeded. Operations above V_c shall be conducted with caution and only in smooth air.

V_a – MANUEVERING SPEED is the maximum speed at which you may use abrupt control travel.

V_{fe} – MAXIMUM FLAP EXTENDED SPEED is the highest speed permissible with wing flaps in a prescribed extended position.

V_{no} – MAXIMUM STRUCTURAL CRUISING SPEED is the speed that should not be exceeded except in smooth air, then only with caution.

SECTION 1 – AIRCRAFT GENERAL DATA

ENGINE Test Aircraft (P/A Demo)

Manufacturer:	_____	Bombardier Rotax
Model:	_____	Rotax 912UL
Type:	_____	See Engine Manual
RPM Full Throttle:	_____	5800 rpm

PROPELLER

Manufacturer:	_____	R.P.M.
Type:	_____	2 Blade Wood
Diameter:	_____	70 inches
Pitch:	_____	Fixed 38 inches

FUEL

Grade:	_____	93 Oct. Auto Gas
Capacity, Total:	_____	18 gal.
Capacity, Useable:	_____	17 gal.

OIL

Grade:	_____	See Engine Manual
Capacity:	_____	3.5 qt.

SECTION 2 – LIMITATIONS

TABLE OF CONTENTS

	PAGE
AIRSPEED LIMITATIONS	2.1
AIRSPEED INDICATOR MARKINGS	2.2
ENGINE LIMITATIONS	2.3
FUEL LIMITATIONS – UNUSABLE	2.3
FUEL GRADE	2.3
AIRCRAFT CATEGORY	2.4
WEIGHT LIMITATIONS	2.4
CENTER OF GRAVITY LIMITATIONS	2.4
FLIGHT LOAD FACTOR LIMITS	2.5
OPERATION LIMITS	2.5
OTHER LIMITATIONS	2.5
FLAPS	2.5
CROSSWIND	2.5
SMOKING	2.5
PLACARDS TO BE DISPLAYED	2.6

SECTION 1 – AIRCRAFT GENERAL DATA, CONTINUED

V_s – STALL SPEED or the minimum steady flight speed at which the aircraft is controllable.

V_{so} – STALL SPEED or the minimum steady flight speed at which the aircraft is controllable in the landing configuration at the most forward center of gravity.

AIRCRAFT PERFORMANCE & FLIGHT PLANNING TERMINOLOGY

USABLE FUEL is the fuel available for flight planning.

UNUSABLE FUEL is the quantity of fuel that can not be safely used in flight.

WEIGHT & BALANCE TERMINOLOGY

For complete Weight & Balance terminology, see Section 6 of this manual.

REFERENCE DATUM is an imaginary vertical plane from which all horizontal distances are measured for balance purposes.

ARM is the horizontal distance from the reference datum to the center of gravity of an item.

MOMENT is the product of the weight of an item multiplied by its arm.

CENTER OF GRAVITY (C.G.) is the point at which an aircraft, or equipment, would balance is suspended. Its distance from the reference datum is found by dividing the total moment by the total weight of the aircraft.

CENTER OF GRAVITY LIMITS are the extreme C.G. locations within which the aircraft must be operated at a given weight.

BASIC EMPTY WEIGHT is the weight of the aircraft including unusable fuel, full operating fluids and full engine oil.

GROSS WEIGHT is the maximum weight to which the aircraft can be loaded for taxi, take-off, flight and landing.

SECTION 2 – LIMITATIONS, CONTINUED

AIRSPEED INDICATOR MARKINGS

	Test Aircraft	(P/A Demo)
White Arc:	_____	40-85 mph
Full Flap Operating Range. Lower limit is at Gross Weight. V _{so} in landing configuration. Upper limit is maximum speed permissible with flaps extended.		
Green Arc:	_____	50-100 mph
Normal Operating Range. Lower limit is at Gross Weight. V _s at most forward C.G. with flaps retracted. Upper limit is maximum structural cruising speed.		
Yellow Arc:	_____	100-115 mph
Operations must be conducted with caution and only in smooth air.		
Red Line:	_____	115 mph
Maximum Speed. Not to be exceeded.		

SECTION 2 – LIMITATIONS

AIRSPEED LIMITATIONS

	Test Aircraft	(P/A Demo)
V _a – Design Maneuvering Speed:	_____	85 mph
V _c – Design Cruising Speed:	_____	85 mph
V _{fe} – Maximum Flap Extended Speed: (Flaps 20°/30°)	_____	80 mph
V _{le} – Maximum Landing Gear Extended Speed:	_____	None
V _{lo} – Maximum Landing Gear Operating Speed:	_____	None
V _{lof} – Lift-off Speed: (Flaps 20°)	_____	52 mph
V _{ne} – Never Exceed Speed:	_____	115 mph
V _{no} Maximum Structural Cruising Speed:	_____	100 mph
V _r – Rotation Speed: (Flaps 20°)	_____	50 mph
V _{so} – Stall Speed in Landing Configuration: (Flaps 30°)	_____	38 mph
V _x – Speed for Best Angle of Climb: (Flaps 20°)	_____	65 mph
V _y – Speed for Best Rate of Climb: (Flaps 10°)	_____	70 mph
Best Speed for Approach: (Flaps 10°/20°)	_____	70 mph
Speed to Raise/Lower Landing Gear in Water: (Engine at Idle)	_____	Dead Slow
Best Approach for Short Field Landing: (Flaps 10°/20°)	_____	62 mph

NOTE: Take-off and landing on water should be done using 20° of flaps under all conditions.

SECTION 2 – LIMITATIONS, CONTINUED

EXPERIMENTAL-TYPE CATEGORY

This aircraft is accepted in the Experimental Type Category Only.

Aerobatic maneuvers including spins are not permitted.

WEIGHT LIMITATIONS

	Test Aircraft	(P/A Demo)
Maximum Take-off Weight:	_____	1370 lbs
Maximum Landing Weight:	_____	1370 lbs
Baggage Compartment Loading Max. Permissible Compartment Load:	_____	55 lbs
Maximum Occupant Loading Maximum Combined Occupant Load:	_____	380 lbs
Minimum Occupant Loading Min. Occupant Load to be not less than:	_____	116 lbs

CENTER OF GRAVITY LIMITATIONS

Center of Gravity Range

Forward Limit Aft of Datum:	_____	97.5 in.
Rear Limit Aft of Datum:	_____	105 in.
Datum, Test Aircraft:	_____	

(P/A Demo Datum used above is 70 inches forward of the wing root leading edge.)

SECTION 2 – LIMITATIONS, CONTINUED

ENGINE LIMITATIONS

Power, Pressure and Temperatures:

	Test Aircraft		(P/A Demo)	
	Minimum	Maximum	Minimum	Maximum
Take-off Power Setting for 5 min.			---	5,800 rpm
Minimum Safe Idling			1,400 rpm	---
Oil Pressure Green Arc 22-58 psi			22 psi Red Line	58 psi Red Line
Oil Temp. Green Arc 130°-240°F			130° F Red Line	280° F Red Line
Cylinder Head Temperatures			---	300° F Red Line

FUEL LIMITATIONS

	Test Aircraft	(P/A Demo)
Take-off/Climb Configuration Unusable Fuel:	_____	.5 Gal.
Cruise Configuration Unusable Fuel:	_____	1.5 Gal.
Landing/Descent Configuration Unusable Fuel:	_____	3 Gal.
Fuel Grades:	_____	See Engine Manual

SECTION 2 – LIMITATIONS, CONTINUED

PLACARDS

The following information must be displayed:

EXPERIMENTAL AMATEUR BUILT AIRCRAFT placard (listing Model, Serial No., Date of Manufacture, Empty Weight, Gross Weight, Engine, Horsepower, and Name and Address of Builder) installed in full view either internally or externally. Recommended installation: on center pylon.

PASSENGER WARNING placard (stating "This aircraft is amateur built and does not comply with the federal safety regulations for standard aircraft"), installed in full view of passenger. Recommended installation: on instrument panel.

EXPERIMENTAL decal installed in full view of passenger. Recommended installation: on lower outboard cockpit tube or floor pan.

DOCUMENTATION

The following documentation must be carried on aircraft and accessible in the cockpit at all times: The Airworthiness Certificate and the Registration. Recommended: waterproof covering.

SECTION 2 – LIMITATIONS, CONTINUED

FLIGHT LOAD FACTOR LIMITS

	Test Aircraft	(P/A Demo)
--	---------------	------------

Normal Category Flight Load Factors:	_____	+3.8g, -1.52g
---	-------	---------------

Avoid coarse control movements above $V_a = 80$ mph to ensure these limit load factors are not exceeded.

KINDS OF OPERATION LIMITS

This aircraft is approved for day VFR and may be equipped for night VFR operations.

OTHER LIMITATIONS

Flap Limitations

Flap setting for take-off and landing, Land & Water, Normal Operations:	_____	20°
--	-------	-----

Crosswind Component

Maximum permissible crosswind component for take-off and landing:	_____	17 mph
--	-------	--------

Smoking

Smoking is not permitted at any time.

SECTION 3 – EMERGENCY PROCEDURES

AIRSPEEDS FOR EMERGENCY OPERATION

	Test Aircraft	(P/A Demo)
Engine Failure After Take-off:		
Wing Flaps, Up	_____	69 mph
Wing Flaps, Down 20°	_____	64 mph

Landing Without Engine

Wing Flaps, Up	_____	69 mph
Wing Flaps, Down 20°	_____	64 mph

ENGINE FAILURES

During Take-off Run

1. Throttle	_____	CLOSE
2. Brakes	_____	APPLY
3. Ignition Switch	_____	OFF

Immediately After Take-off

1. Airspeed – Flaps UP	_____	69 mph
Airspeed – Flaps DOWN	_____	64 mph
2. Fuel	_____	OFF
3. Ignition Switch	_____	OFF
4. Wing Flaps	_____	AS REQ'D

Other Procedures:

- If sufficient runway/strip/waterway is available, lower the nose sufficiently to maintain speed and make a normal landing straight ahead.
- If over the airport boundary and/or in a position where obstacles are in the path of the aircraft and height permits, a slight change in heading may be made to line up on the most suitable landing area.
- If time permits, carry out the engine failure checks listed on Page 3.2 – *Initial Checks*.

During Flight

1. Airspeed	_____	69 mph
2. Fuel (Troubleshoot)	_____	CHECK ON
3. Electric Fuel Pump, if installed	_____	ON
4. Ignition Switch – If Propeller is stopped	_____	START

SECTION 3 – EMERGENCY PROCEDURES

TABLE OF CONTENTS

	PAGE
AIRSPEEDS FOR EMERGENCY OPERATION	3.1
ENGINE FAILURE	3.1
During Take-off Run	3.1
Immediately After Take-off	3.1
During Flight	3.1
Restarting Engine in Flight	3.2
Engine Failure Checks	3.2
FORCED LANDINGS	3.3
Emergency Landing Without Engine Power	3.3
Precautionary Landing With Engine Power	3.4
Ditching	3.4
FIRES	3.5
Action In the Event of Fire	3.5
Engine Fire on the Ground	3.5
Engine Fire in Flight	3.5
Electrical Fire in Flight	3.5
ICING	3.6
Inadvertent Icing Encounter	3.6
ELECTRICAL POWER SYSTEM MALFUNCTIONS	3.6
Ammeter Shows Excessive Rate of Charge	3.6
Generator “Out” Light Illuminates During Flight	3.6
ROUGH ENGINE OR LOSS OF POWER	3.7
Carburetor Icing	3.7
Spark Plug Fouling	3.7
Low Oil Pressure	3.7
SPINS	3.7

SECTION 3 – EMERGENCY PROCEDURES, CONTINUED

FORCED LANDINGS

Emergency Landing Without Engine Power

	Test Aircraft	(P/A Demo)
1. Airspeed	_____	69 mph
2. Fuel	_____	OFF
3. Ignition Switch	_____	OFF
4. Radio, if available	_____	Transmit Distress Call
5. Transponder, if available	_____	Set code 7700
6. Wing Flaps	_____	20°
7. Sliding Canopies	_____	Unlatched, Partially Open
8. Touchdown	_____	LAND-Brake hard WATER – As normal, if short, back stick

Other procedures:

- a. Convert any excess speed to height and at the same time perform the checks listed on Page 3.2, *Initial Checks*.
- b. Place the aircraft at optimum gliding angle/attitude/speed.
- c. Select the most suitable field/waterway, bearing in mind height above the ground, wind speed and direction and the availability of assistance after landing.
- d. Plan descent in relation to the selected field/waterway – aiming to reach a base leg position – as for a normal glide approach.
- e. Keeping within easy gliding distance of the field/waterway, carry out the *Troubleshoot* on Page 3.2, *Initial Checks*.
- f. If the engine cannot be restarted, continue as planned, brief the passenger and send distress signal, if radio is available.
- g. During descent, use flaps intelligently. Aim to have 10° of flap on base leg – use 20° when you are sure of making the field/waterway – turn off all fuel and electrical switches at a time early enough not to interfere with concentration over the final stages of the emergency landing. Refer to Page 3.2, *Safety Check*.

Note: If intending to use the aircraft radio for communication, make a thorough inspection to ensure that no fuel is spilled that would be likely to cause a fire if the electrical circuits are reactivated.

SECTION 3 – EMERGENCY PROCEDURES, CONTINUED

Restarting The Engine in Flight

General – It is unlikely that the engine will stop during normal maneuvers. However, in the event of the engine stopping in flight the following procedures will normally allow the engine to be restarted without problems.

Using Starter Motor

	Test Aircraft	(P/A Demo)
1. Throttle	_____	OPEN ½"
2. Fuel	_____	CHECK ON
3. Electric Fuel Pump, if Installed	_____	ON
4. Master Switch	_____	ON
5. Ignition	_____	BOTH L & R ON
5. Starter	_____	Turn Key & Start

CAUTION

Because of the high compression ratio of the Rotax four-cycle engines, the propeller will not windmill, even in a steep dive. If the engine cannot be restarted by using the starter motor, carry out the procedure as set out in "*Emergency Landing Without Power*" on Page 3.3.

Engine Failure Checks

Initial Check

Fuel ON – Check contents – Fuel filter bowl, if installed.

Troubleshoot

1. Fuel ON – Contents sufficient
2. Oil pressure & temperature normal
3. Switches – Both ON – Check L & R Ignition
4. Throttle checked for operation – Open

Safety Check

1. Brakes OFF
2. Switches, Ignition and Master OFF
3. Fuel OFF
4. Harness secure – All occupants
5. Canopies/Hatches set for landing
6. Undercarriage – set for landing

SECTION 3 – EMERGENCY PROCEDURES, CONTINUED

FIRE

Action in the Event of Fire

General – due to the high octane fuel carried by the aircraft, there is always the possibility of a violent explosion occurring if an aircraft catches fire. Personnel are therefore warned not to take undue risks in attempting to save an aircraft which is obviously well alight. A small fire extinguisher is only meant to be used on small fires. For large fires, trained personnel with appropriate equipment are required.

Engine Fire on the Ground

1. Throttle CLOSED
2. Ignition OFF
3. Fuel OFF
4. Master Switch OFF
5. Evacuate the aircraft
6. Use fire extinguishers as applicable.

Engine Fire In Flight

Engine Fire – Symptoms confirmed – Shut Down Immediately

1. Throttle CLOSED
2. Ignition OFF
3. Fuel OFF
4. Landing Plan EMERGENCY descent and landing
5. Radio, if available TRANSMIT MAYDAY CALL
6. Emergency landing Refer to Page 3.2, *Safety Check*

Electrical Fire In Flight

1. Master Switch OFF
2. Troubleshoot Identify and Isolate Faulty Circuit
3. Canopy CLOSED
4. Fire Extinguisher ACTIVATE (if available)
WARNING: After discharging extinguisher within closed cabin, open sliding canopies to ventilate cabin.
5. Carry out an emergency landing at the nearest appropriate airport/waterway or landing ground.
6. If the fire persists, sideslip away from the flames. Instruct the passenger as necessary and proceed to an early landing.

Electrical Power

Do not turn the generator (if installed) off in flight, except in an emergency.

SECTION 3 – EMERGENCY PROCEDURES, CONTINUED

Precautionary Landing With Engine Power

In the event of a landing having to be made by virtue of deteriorating weather conditions, or for any reason where it is impossible to continue a flight and where no engine malfunction has occurred, proceed as follows:

	Test Aircraft	(P/A Demo)
1. Airspeed	_____	69 mph
2. Wing Flaps	_____	20°

Other Procedures:

- a. Aim to land the aircraft at the nearest airfield/waterway or authorized landing ground, suitable for the type.
- b. If worsening flying conditions make this impossible or a proper landing ground is unavailable, select the largest and best open area for landing (as close as possible to habitation).
- c. Where any area other than a recognized airport or waterway is to be used, make dummy runs over the selected approach and landing path. The dummy runs should be flown at various heights where possible. Never carry out dummy runs up and down sloping terrain, always fly across the slope.
- d. Where appropriate, carry out a short field landing.

Ditching

1. Landing Gear: UP POSITION
2. Radio, if available: TRANSMIT DISTRESS CALL
3. Transponder, if available: SQUAWK 7700
4. Baggage, etc.: SECURE OR JETTISON
5. Approach: HIGH WINDS/HEAVY SEAS – Into Wind
 LIGHT WINDS/HEAVY SWELL – Parallel to swell
6. Wing Flaps: 20°
7. Power: 100 FT/MIN DESCENT AT 60 MPH
8. Passenger: Brief – Headset, Seat Belts, Canopy, Orientation, Cockpit Exit, Life Jacket/Raft, Face Protection, Clothing

SECTION 3 – EMERGENCY PROCEDURES, CONTINUED

ROUGH ENGINE OPERATION OR LOSS OF POWER

Carburetor Icing

A gradual loss of RPM and eventual engine roughness may result from the formation of carburetor ice. If carburetor heat is available: To clear the ice, apply full throttle and pull the carburetor heat knob full out until the engine runs smoothly. Then remove the carburetor heat and adjust the throttle.

Spark Plug Fouling

A slight engine roughness in flight may be caused by one or more spark plugs becoming fouled. This may be verified by turning the ignition switch from BOTH to either L or R position. An obvious power loss in a single ignition operation is evidence of spark plug trouble.

Low Oil Pressure

If low oil pressure is accompanied by normal oil temperature, there is a possibility the oil pressure gauge or relief valve is malfunctioning. If the condition persists a landing at the nearest airport would be advisable to inspect the source of trouble.

If a total loss of oil pressure is accompanied by a rise in oil temperature, there is every reason to suspect an engine failure is imminent. Reduce engine power immediately and select a suitable forced landing field, using only the minimum amount of power application to carry out the landing.

SPINS

DELIBERATE SPINS ARE PROHIBITED

If an inadvertent spin is encountered, proceed as follows:

- | | |
|-------------------------|---|
| 1. Throttle | CLOSED |
| 2. Ailerons | NEUTRAL |
| 3. Spin Direction | IDENTIFY |
| 4. Rudder | FULL OPPOSITE to direction of spin |
| 5. Stick | Progressively forward until rotation ceases |
| 6. When rotation ceases | Level wings and recover from dive |

SECTION 3 – EMERGENCY PROCEDURES, CONTINUED

ICING

Inadvertent Icing Encounter

1. Turn pitot heat switch ON if installed.
2. Alter course or change altitude to obtain an outside air temperature that is less conducive to icing.
3. Open throttle to increase engine speed and minimize ice buildup on propeller blades.
4. Watch for signs of carburetor icing and apply carburetor heat, if installed, as necessary. Loss in engine RPM could be caused by carburetor ice buildup.
5. If airframe ice formation is rapid or buildup is significant, plan a landing at the nearest airport.
6. Be prepared for a higher stall speed.
7. With ice buildup, make faster landing approach, 69-75 mph.

ELECTRICAL POWER SUPPLY SYSTEM MALFUNCTIONS

Ammeter Shows Excessive Rate of Charge

- | | |
|-----------------------------------|-----------|
| 1. Generator | OFF |
| 2. Generator C/Breaker | PULL OFF |
| 3. Avionics & High Load Equipment | OFF |
| 4. Flight | TERMINATE |

Generator “Out” Light (if fitted) Illuminates During Flight – Ammeter Indicates Discharge

- | | |
|----------------------------|---------------|
| 1. Non-essential Equipment | OFF |
| 2. Generator C/Breaker | CHECK – IN |
| 3. Master Switch | OFF |
| 4. Master Switch | ON |
| 5. Low Voltage Light | CHECK OFF |
| 6. Avionics & Equipment | TURN ON AGAIN |

If low voltage light illuminates again:

- | | |
|--------------------------------|-----------|
| 7. Generator | OFF |
| 8. All Non-essential Equipment | OFF |
| 9. Flight | TERMINATE |

SECTION 4 – NORMAL PROCEDURES

This Section contains essential information relating to performance and handling characteristics for conducting normal operation of the aircraft.

SPEEDS FOR NORMAL OPERATIONS

	Test Aircraft	(P/A Demo)
Va – Design Maneuvering Speed:	_____	85 mph
Vc – Design Cruising Speed:	_____	85 mph
Vfe – Maximum Flap Extended Speed: (Flaps 20°/30°)	_____	80 mph
Vle – Maximum Landing Gear Extended Speed:	_____	None
Vlo – Maximum Landing Gear Operating Speed:	_____	None
Vlof – Lift-off Speed: (Flaps 20°)	_____	52 mph
Vne – Never Exceed Speed:	_____	115 mph
Vno Maximum Structural Cruising Speed:	_____	100 mph
Vr – Rotation Speed (Flaps 20°):	_____	50 mph
Vso – Stall Speed in Landing Configuration: (Flaps 30°)	_____	38 mph
Vx – Speed for Best Angle of Climb (Flaps 20°):	_____	65 mph
Vy – Speed for Best Rate of Climb (Flaps 10°):	_____	70 mph
Best Speed for Approach (Flaps 10°/20°):	_____	70 mph
Speed to Raise/Lower Landing Gear in Water: (Engine at Idle)	_____	Dead Slow
Best Approach for Short Field Landing: (Flaps 10°/20°)	_____	62 mph

NOTE: Take-off and landing on water should be done using 20° of flaps under normal conditions.

SECTION 4 – NORMAL PROCEDURES

TABLE OF CONTENTS

	PAGE
SPEEDS FOR NORMAL OPERATIONS	4.1
AIRCRAFT SKETCH – PREFLIGHT INSPECTION	4.2
DAILY PREFLIGHT INSPECTION	4.3
DAILY PREFLIGHT INSPECTION, CONTINUED	4.4
COCKPIT SECURITY	4.5
BEFORE STARTING ENGINE	4.5
STARTING ENGINE	4.5
BEFORE TAKE-OFF	4.6
BEFORE TAKE-OFF – WATER OPERATIONS	4.6
TAKE-OFF	4.6
ENROUTE CLIMB	4.6
CRUISE	4.6
DESCENT	4.7
BEFORE LANDING	4.7
LANDING – LAND OPERATIONS	4.7
LANDING – WATER OPERATIONS	4.7
AFTER LANDING	4.7
AFTER LANDING – WATER OPERATIONS	4.8
SHUTDOWN	4.8
SECURING AIRCRAFT	4.8

SECTION 4 – NORMAL PROCEDURES, CONTINUED

DAILY PREFLIGHT INSPECTION

1. Cockpit & Forward Fuselage

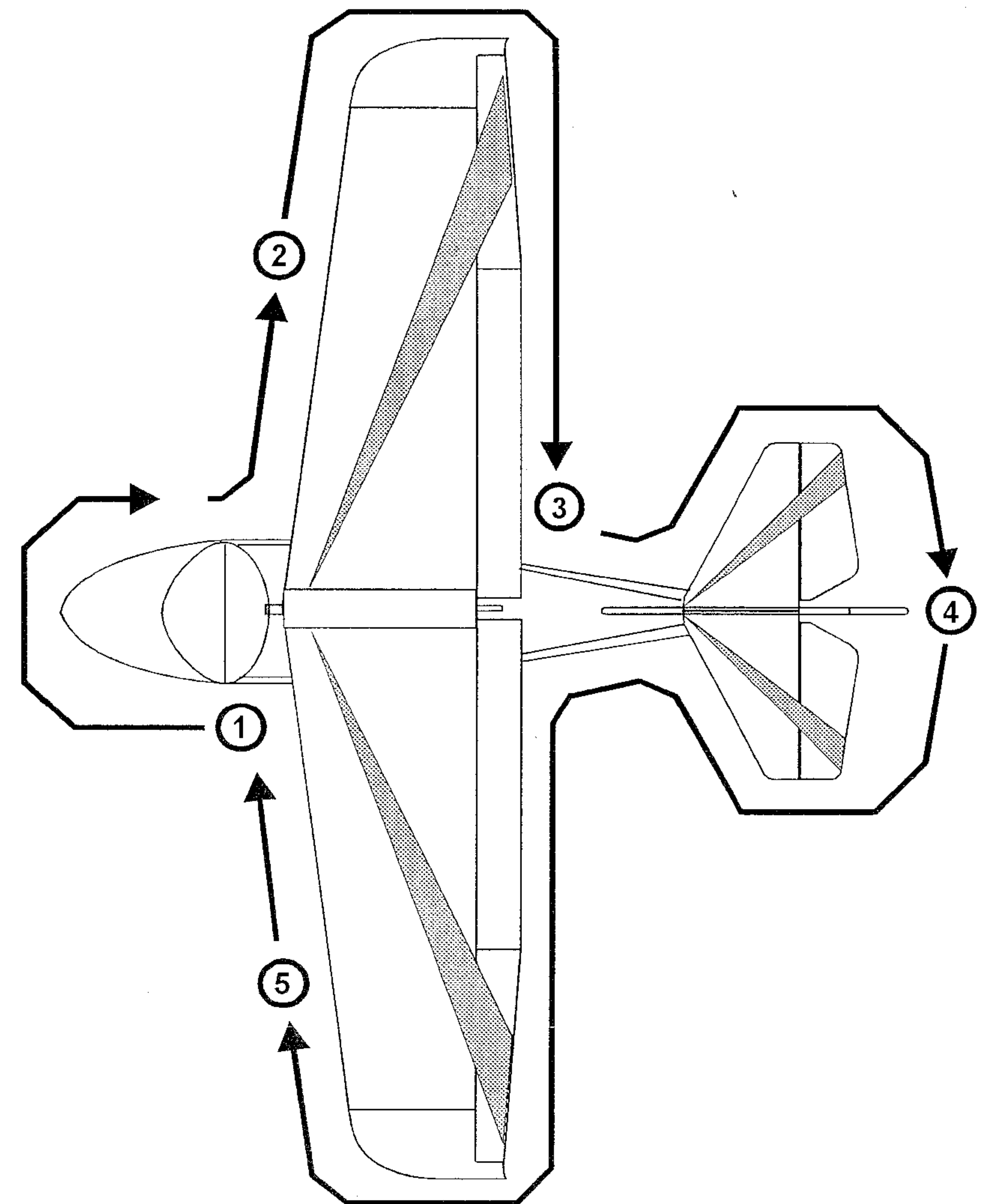
	Test Aircraft	(P/A Demo)
1. Flight Manual & Aircraft Documentation	_____	AVAILABLE
2. Control Lock (if fitted)	_____	REMOVE
3. Ignition Switch	_____	OFF
4. Master Switch	_____	ON
5. Fuel Quantity Visual Quantity & Gauge	_____	CHECK
6. Electric Trim Operation	_____	CHECK
7. Bilge Pump Operation & Hull Water - Drain Any Accumulated Water	_____	CHECK
8. Master Switch	_____	OFF
9. Carburetor Heat Actuation (if fitted)	_____	FREE
10. Undercarriage Overcenter	_____	LOCKED
11. Flight Control Full & Free Movement	_____	CHECK
12. Aileron Push Rods, Bolt Ends & Cables	_____	SECURE
13. Fuel Filter/Glass Bowl-Leaks, Contamination	_____	CHECK
14. Check Fuel for Contamination if Fitted with Fuel Drain or Gascolator	_____	N/A
15. Instruments Free of Damage & Secure	_____	CHECK
16. Static Ports Clear of Blockage & Obstruction	_____	CHECK
17. Windshield Cracks & Clean	_____	CHECK
18. Seat Belts, Inertia Reels Secure & Functional	_____	CHECK
19. Hull Sides & Underside Free of Damage	_____	CHECK

2. Starboard Wing

1. Starboard Tire Inflation, Condition, Attachment	_____	CHECK
2. Lower Strut/Fuselage Attachment Bolts	_____	SECURE
3. Outer Strut & Jury Strut Attachment	_____	SECURE
4. Wing Float, Braces & Secure Attachment	_____	CHECK
5. Aileron Outer Push-Pull Rod Attachment	_____	SECURE
6. Aileron Movement	_____	FREE
7. Flap & Push-Pull Attachment	_____	SECURE
8. Wing to Aft Fuselage Brace Cable	_____	SECURE
9. Wing Tie-Down	_____	REMOVE

SECTION 4 – NORMAL PROCEDURES, CONTINUED

AIRCRAFT SKETCH – PREFLIGHT INSPECTION



SECTION 4 – NORMAL PROCEDURES, CONTINUED

COCKPIT SECURITY

Before starting the engine and conducting any operation, ensure that all articles and equipment are stowed safely and secured.

This is of particular importance in the event of any object exiting the cockpit, which could cause damage to the propeller because of the pusher configuration of this aircraft.

BEFORE STARTING ENGINE

Test Aircraft (P/A Demo)

- 1. Preflight Inspection _____ Complete
- 2. Pull Through Prop – Minimum 2 Revolutions _____ Complete
- 3. Seat Belts & Shoulder Harnesses Adjusted _____ CHECK
- 4. Fuel Selector Valve _____ ON
- 5. All Radio Equipment & Avionics Power _____ OFF
CAUTION – Damage to avionics equipment can occur during engine start if avionics are left on.
- 6. Brakes or Chocks as Required _____ SET
- 7. Circuit Breakers In _____ CHECK

STARTING ENGINE

- 1. Master Switch _____ ON
- 2. Prime (none if engine warm) _____ AS REQD
- 3. Electric Fuel Pump, if Fitted _____ ON
- 4. Throttle ½" approx. _____ OPEN
- 5. Propeller Area _____ CLEAR
- 6. Ignition Switch to Both Position, then _____ START
- 7. Oil Pressure _____ CHECK
- 8. Radio & Electrical Equipment as Required _____ ON

SECTION 4 – NORMAL PROCEDURES, CONTINUED

DAILY PREFLIGHT INSPECTION, CONTINUED

3. Engine

Test Aircraft (P/A Demo)

- 1. Oil Level (Run engine 3 min prior to checking) _____ CHECK
- 2. Propeller – Nicks, Cracks & Security _____ CHECK
- 3. Carburetor Attachment & Induction System _____ SECURE
- 4. All Pipes & Hoses – Leakage, Wear & Security _____ CHECK
- 5. Exhaust & Muffler Springs, Cracks, Attachment _____ CHECK
- 6. Coolant Level & Color _____ CHECK
- 7. Engine Mounts _____ SECURE
- 8. Electrical Cables – Wear & Security _____ CHECK

4. Empennage

- 1. Stabilizer Leading Edge Trim Attachment _____ CHECK
- 2. Elevator Push-Pull Tubes & Horn Attachment _____ CHECK
- 3. Elevators & Rudder – Free Movement, Security _____ CHECK
- 4. Upper & Lower Tail Cables _____ SECURE
- 5. Tail Wheel for Inflation & Wear _____ CHECK
- 6. Tail Tie-Down _____ REMOVE

5. Port Wing

In addition to carrying out checks as for the Starboard Wing:

- 1. Fuel Cap Securely Attached _____ CHECK
- 2. Pitot Tube Clear of Obstruction-Cover Removed _____ CHECK

SECTION 4 – NORMAL PROCEDURES, CONTINUED

DESCENT

Test Aircraft (P/A Demo)

- | | | | |
|----|----------------------------|-------|---------|
| 1. | Fuel Contents | _____ | CHECK |
| 2. | Throttle Set | _____ | AS REQD |
| 3. | Carb Heat to Prevent Icing | _____ | AS REQD |

BEFORE LANDING – DOWN WIND CHECK

- | | | | |
|----|----------------------------------|-------|-------------|
| 1. | Canopies Closed but Not Locked | _____ | CHECK |
| 2. | Seat Belts & Shoulder Harnesses | _____ | SECURE |
| 3. | Carburetor Heat (if fitted) | _____ | AS REQD |
| 4. | Fuel Contents | _____ | CHECK |
| 5. | Fuel Pump | _____ | ON |
| 6. | Undercarriage for GROUND LANDING | _____ | DOWN |
| | Over Center Lock | _____ | CHECK |
| | Undercarriage for WATER LANDING | _____ | UP |
| | Over Center Lock | _____ | CHECK |
| 7. | Flaps Set | _____ | 10° |

LANDING – LAND OPERATIONS

- | | | | |
|----|-------------------------------------|-------|-------------|
| 1. | Airspeed Approach | _____ | 65-75 mph |
| 2. | Flaps Set | _____ | 20° |
| 3. | Undercarriage – DOUBLE CHECK | _____ | DOWN |
| 4. | Touchdown | _____ | 46-52 mph |

LANDING – WATER OPERATIONS

- | | | | |
|----|---|-------|-----------|
| 1. | Airspeed Approach | _____ | 65-75 mph |
| 2. | Flaps Set | _____ | 20° |
| 3. | Undercarriage – DOUBLE CHECK | _____ | UP |
| 4. | Canopies Closed But Not Locked | _____ | CHECK |
| 5. | Touchdown – CARE in Glassy Water Conditions | _____ | 46-52 mph |

AFTER LANDING

- | | | | |
|----|---------------|-------|---------|
| 1. | Wing Flaps | _____ | UP |
| 2. | Fuel Pump | _____ | OFF |
| 3. | Strobe Lights | _____ | AS REQD |

SECTION 4 – NORMAL PROCEDURES, CONTINUED

BEFORE TAKE-OFF

Test Aircraft (P/A Demo)

- | | | | |
|-----|---|-------|----------|
| 1. | Hatches as Required
(closed but not locked during water operation) | _____ | CHECK |
| 2. | Seat Belts & Shoulder Harness On | _____ | SECURE |
| 3. | Trim – Full Up for Take-off | _____ | SET |
| 4. | Fuel – Double Check | _____ | ON |
| 5. | Electric Fuel Pump, if Fitted | _____ | ON |
| 6. | Flaps - 20° | _____ | SET |
| 7. | Engine Run Up 3500 RPM | _____ | COMPLETE |
| | L & R Ignition - RPM drop-not less than 300 | _____ | CHECK |
| | Carb Heat, if fitted – note RPM drop | _____ | CHECK |
| | Engine Instruments (Temps) & Ammeter | _____ | CHECK |
| | Minimum Oil Temperature 130 ° F | _____ | CHECK |
| 8. | Throttle Reduced to Idle | _____ | 1700 RPM |
| 9. | Radios Frequencies etc., if Fitted | _____ | SET |
| 10. | Transponder (if fitted) | _____ | SET |
| 11. | Strobe Lights (if fitted) | _____ | ON |
| 12. | Clearance | _____ | AS REQD |

BEFORE TAKE-OFF – (ADDITION FOR WATER OPERATION)

- | | | | |
|----|---------------------------|-------|--------|
| 1. | Undercarriage UP & LOCKED | _____ | CHECK |
| 2. | Bilge Pump | _____ | ON/OFF |

TAKE-OFF

- | | | | |
|----|--|-------|-----------|
| 1. | Throttle Maximum 5800 RPM | _____ | OPEN |
| 2. | LAND ONLY: Elevator Control Forward to Lift Tail | _____ | AS REQD |
| 3. | Lift Off | _____ | 46-52 mph |
| 4. | Accelerate to | _____ | 65 mph |
| 5. | Climb Out and When Established
Retract Undercarriage (Ground Operation) | _____ | COMPLETE |
| 6. | At Safe Height Reduce Flaps to 10° | _____ | |

ENROUTE CLIMB

- | | | | |
|----|-----------|-------|----------|
| 1. | Airspeed | _____ | 63 mph |
| 2. | Throttle | _____ | 5500 RPM |
| 3. | Flaps Set | _____ | UP |
| 4. | Fuel Pump | _____ | OFF |

CRUISE

- | | | | |
|----|----------------------|-------|----------------|
| 1. | Throttle As Required | _____ | 4900 –5200 rpm |
| 2. | Speed | _____ | 80-90 mph |
| 3. | Trim | _____ | AS REQD |

SECTION 5 – PERFORMANCE

TABLE OF CONTENTS

	PAGE
PERFORMANCE CHARTS – INTRODUCTION	5.1
TAKE-OFF PERFORMANCE – GENERAL	5.1
TAKE-OFF PERFORMANCE CHART	5.2
SAMPLE TAKE-OFF PERFORMANCE CHART	5.3
CLIMB PERFORMANCE CHART	5.4
LANDING PERFORMANCE – GENERAL	5.5
LANDING PERFORMANCE CHART	5.6
SAMPLE LANDING PERFORMANCE CHART	5.7

SECTION 4 – NORMAL PROCEDURES, CONTINUED

AFTER LANDING – Water Operations – Docking & Beaching

In the case of Docking or Beaching the following points are recommended to assist in effective & safe operation.

	Test Aircraft (P/A Demo)
1. Radio	_____ OFF
2. Headsets	_____ REMOVE
3. Seat Belts & Harnesses	_____ RELEASE
4. Undercarriage DOWN/UP & LOCKED	_____ AS REQD
5. Canopies	_____ OPEN
6. Ropes	_____ READY
7. Ignition – When Required	_____ OFF
8. Master Switch	_____ OFF

SHUT DOWN

1. Radio	_____ OFF
2. All Electrical Equipment	_____ OFF
3. Ignition Switch	_____ OFF
4. Master Switch	_____ OFF

SECURING AIRCRAFT

1. All Radio & Electrical Equipment	_____ OFF
2. Ignition Switch	_____ OFF
3. Master Switches	_____ OFF
4. Controls Tethered with Seat Belt	_____ SECURE
5. Canopies	_____ LOCKED
6. Pitot Tube Cover	_____ FITTED
7. Chocks	_____ LOCATED
8. Tie-Downs	_____ SECURE
9. Cockpit/Engine Covers Fitted	_____ AS REQD

SECTION 5 – PERFORMANCE, CONTINUED

TAKE-OFF PERFORMANCE CHART – TEST AIRCRAFT

Pressure Altitude – 0 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

Pressure Altitude – 2,500 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

Pressure Altitude – 5,000 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

Pressure Altitude – 7,500 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

SECTION 5 – PERFORMANCE

PERFORMANCE CHARTS – INTRODUCTION

The charts in this Section contain data establishing runway and waterway lengths for take-off, landing and climb performance at a gross weight of 1370 lbs.

TAKE-OFF PERFORMANCE – GENERAL

The maximum gross weight for take-off shall not exceed the maximum take-off weight specified in Section 2 of this Manual.

The take-off distance (ground run in feet) with full throttle, flaps deflected 20° and a take-off safety speed of 50 mph can be determined from the following chart. The take-off distances are for a hard surface runway and/or waterway. Soft ground and/or wet grass will increase the land take-off distance and pilots should satisfy themselves that adequate runway is available to cover these conditions.

The take-off distance can be read from the table, at the appropriate values of pressure altitude and outside air temperature. For values of temperature and altitude not listed, interpolation between the values in the table is permitted.

Ground Take-off: Before commencing the take-off roll, check that the gear is in the fully “down” and locked position. To do this check that the over center lock indicator is in the full forward position. After take-off retract the gear only after the take-off safety speed is reached and the aircraft is established in the climb.

Water Take-off: The aircraft may be taxied slowly in the water with the gear in the down position, when transitioning from land or water. Check that the gear is retracted before commencing the take-off run.

SECTION 5 – PERFORMANCE, CONTINUED

CLIMB PERFORMANCE

The climb performance can be determined from the following chart. This chart assumes that maximum take-off power is used.

CLIMB PERFORMANCE CHART

Altitude Feet	Test Aircraft	(P/A Demo)
	ROC f.p.m.	ROC f.p.m.
0	_____	400
1,000	_____	373
2,000	_____	348
3,000	_____	323
4,000	_____	299
5,000	_____	276
6,000	_____	254
7,000	_____	233
8,000	_____	212
9,000	_____	192
10,000	_____	173
11,000	_____	154
12,000	_____	137
13,000	_____	120
14,000	_____	104

Note: Climb Data is for Standard Atmospheric Conditions.
i.e. 59° F and 29.92 H.g.

SECTION 5 – PERFORMANCE, CONTINUED

SAMPLE TAKE-OFF PERFORMANCE CHART – (P/A DEMO)

Pressure Altitude – 0 Feet						
	59° F. OAT			86° F. OAT		
	0	10	20	0	10	20
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	522	312	155	546	326	163
Distance to 50 Feet	1078	742	459	1128	776	480

Pressure Altitude – 2,500 Feet						
	59° F. OAT			86° F. OAT		
	0	10	20	0	10	20
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	683	408	203	714	427	213
Distance to 50 Feet	1411	970	600	1476	1015	628

Pressure Altitude – 5,000 Feet						
	59° F. OAT			86° F. OAT		
	0	10	20	0	10	20
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	906	541	270	948	566	282
Distance to 50 Feet	1872	1287	796	1958	1347	833

Pressure Altitude – 7,500 Feet						
	59° F. OAT			86° F. OAT		
	0	10	20	0	10	20
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	1237	739	368	1294	773	385
Distance to 50 Feet	2555	1757	1087	2673	1839	1137

SECTION 5 – PERFORMANCE, CONTINUED

LANDING PERFORMANCE – GENERAL

	Test Aircraft	(P/A Demo)
Minimum Approach Speed 20° Flap	_____	58 mph
Normal Approach Speed: 20° Flap	_____	63 – 69 mph

The landing distances (ground run in feet) that appear in the chart have been calculated by using the gross weight of 1370 lbs.

The landing distance can be read from the table, at the appropriate values of pressure altitude and outside air temperature. For values of temperature and altitude not listed, interpolation between the values in the table is permitted.

These distances are derived using the above minimum approach speed with 20° of flap and engine at idle. After touchdown maximum braking is used to bring the aircraft to a stop. These distances are for a hard level surface. Wet and/or slippery surfaces will increase these distances and pilots should satisfy themselves that adequate runway length is available to cover these conditions.

In the case of water landings, full reduction of power after touchdown and application of full up elevator below 25 mph will result in best speed reduction and reduced landing distance covered.

Ground Landing: Check that the gear is fully extended in the “down” and locked position. To do this, check that the over center lock indicator is in the full forward position. In the event that the gear cannot be extended, use a grass runway and land the aircraft on the hull. Minimal damage will result.

Water Landing: Check that the gear is fully retracted and locked in the “up” position.

Note: In the interest of safety and good airmanship, pilots should include in the pre-landing check schedule, the habit of double-checking the undercarriage for correct position, particularly when carrying out water landings. This practice is of particular importance.

SECTION 5 – PERFORMANCE, CONTINUED

CLIMB PERFORMANCE

The climb performance can be determined from the following chart. This chart assumes that maximum take-off power is used.

CLIMB PERFORMANCE CHART

Altitude Feet	Test Aircraft ROC f.p.m.	(P/A Demo) ROC f.p.m.
0	_____	400
1,000	_____	373
2,000	_____	348
3,000	_____	323
4,000	_____	299
5,000	_____	276
6,000	_____	254
7,000	_____	233
8,000	_____	212
9,000	_____	192
10,000	_____	173
11,000	_____	154
12,000	_____	137
13,000	_____	120
14,000	_____	104

Note: Climb Data is for Standard Atmospheric Conditions. i.e. 59° F and 29.92 H.g.

SECTION 5 – PERFORMANCE, CONTINUED

SAMPLE LANDING PERFORMANCE CHART – (P/A DEMO)

Pressure Altitude – 0 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	655	438	265	685	459	277
Distance to 50 Feet	1492	1123	798	1522	1144	810

Pressure Altitude – 2,500 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	718	480	290	751	503	304
Distance to 50 Feet	1555	1165	823	1588	1187	836

Pressure Altitude – 5,000 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	787	527	319	824	551	333
Distance to 50 Feet	1624	1212	851	1661	1236	866

Pressure Altitude – 7,500 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet	856	579	350	905	606	366
Distance to 50 Feet	1702	1264	883	1742	1291	899

SECTION 5 – PERFORMANCE, CONTINUED

LANDING PERFORMANCE CHART – TEST AIRCRAFT

Pressure Altitude – 0 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

Pressure Altitude – 2,500 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

Pressure Altitude – 5,000 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

Pressure Altitude – 7,500 Feet						
	59° F. OAT			86° F. OAT		
Wind – mph	0	10	20	0	10	20
Ground Roll – Feet						
Distance to 50 Feet						

SECTION 6 – WEIGHT AND BALANCE (LOADING)

WEIGHT AND BALANCE INFORMATION

All aircraft are structurally and aerodynamically engineered for certain load conditions which result from specific weights and forces anticipated to occur in normal operations within its specified flight envelope. An aircraft's handling qualities and structural integrity may be seriously compromised if the weight and balance limits are exceeded in normal operations.

It is the pilot's responsibility to make sure the weight and balance limits are not exceeded as to weight, its location, distribution and security prior to any flight.

DEFINITIONS

EMPTY WEIGHT: The actual weight of the individual aircraft, including the structure, power plant, fixed equipment, any fixed ballast, unusable (in-flight) fuel, lubricants and coolant.

Original Empty Weight is determined by actually weighing the new aircraft before it is flown.

Any time a major alteration, modification, or repair is performed on the aircraft, its new Empty Weight must be determined by either weighing the aircraft again, or by accurate calculation of the weight changes and their effect on Empty Weight Center of Gravity (EWCG) location.

A major alteration or modification results from the addition, deletion, or redistribution of existing equipment and accessories, or from a repair which results in a significant increase of weight of the airframe or engine. For example, addition or removal of battery, radios, installation of a larger fuel tank or engine, painting the airframe, installation of heavier wheels and tires, etc.

GROSS WEIGHT: The maximum total weight for which an aircraft's structure and performance have been approved for normal operations by its manufacturer. It is the maximum weight (Empty Weight plus useful load) at which an aircraft can be safely operated. Maximum take-off weight must never exceed the published Gross Weight.

USEFUL LOAD: The total amount of weight available for pilot, passengers, baggage, cargo and in-flight usable fuel.

MAXIMUM/MINIMUM WEIGHTS: Due to certain balance, structural and aerodynamic considerations, sometimes a maximum or minimum weight may be specified for certain locations on the aircraft.

CENTER OF GRAVITY (C.G.): A point along an aircraft's longitudinal axis at which all the loads and forces are perfectly concentrated and balanced.

SECTION 6 – WEIGHT AND BALANCE

TABLE OF CONTENTS

	PAGE
WEIGHT AND BALANCE INFORMATION	6.1
DEFINITIONS	6.1
DEFINITIONS, CONTINUED	6.2
PROCEDURE	6.3
EMPTY WEIGHT CENTER OF GRAVITY CALCULATION	6.4
LOADED WEIGHT AND BALANCE CALCULATIONS	6.4
CRITICAL LOADING CONDITIONS	6.5
LIST OF INSTALLED EQUIPMENT	6.6
WEIGHT AND BALANCE FORM	6.7
WEIGHT AND BALANCE LIMITS	6.7
WEIGHT AND BALANCE DATA SHEET	6.8
EMPTY WEIGHT AND BALANCE FORM	6.8

SECTION 6 – WEIGHT AND BALANCE, CONTINUED

PROCEDURE

All permanent equipment, options and accessories should be installed on the aircraft prior to weighing. All equipment options and accessories installed in the aircraft must be listed on the "installed Equipment List". That list becomes part of Weight and Balance Documents.

Be sure to remove any loose equipment, tools, etc. from the aircraft prior to weighing.

The fuel tank should be empty except for unusable fuel. If the fuel tank is not empty, then the exact amount of usable fuel in the tank must be determined. Usable fuel weight and its moment must be deducted from the empty weight calculations before E.W.C.G. can be accurately determined.

Oil and coolant tanks and reservoirs must be properly filled before weighing. These and any other liquids necessary for normal operations are considered part of an aircraft's empty weight.

If weighing is done outdoors, make sure there is no wind to affect the weight measurements. For best results, weigh indoors.

The scales must be calibrated correctly. All scales must be set on level ground.

Any equipment placed on the scales when weighing the aircraft, such as chocks or blocks, should be weighed separately and the weight deducted from the scale reading. These weights should be noted for reference, if necessary.

The aircraft must be weighed in a level flight attitude, both longitudinally (front to back) and laterally, as shown in the Weight and Balance Data Sheet.

Place a scale under each wheel of the aircraft. Record the weight of each scale on the "Empty Weight and Balance Form", as shown in the Weight and Balance Data Sheet.

Measure the exact horizontal distance from the datum line to center of spindles of wheel axles, as shown in Figure 2. Record these measurements on the Empty Weight and Balance Form.

If only one scale is used for weighing, be sure to level the wheels not being weighed before taking the scale readings. Remember, the aircraft must be in proper level flight attitude to ensure accuracy.

SECTION 6 – WEIGHT AND BALANCE, CONTINUED

CENTER OF GRAVITY RANGE: The horizontal distance, along an aircraft's longitudinal axis, within which an aircraft has been found to be fully maneuverable at all specified design speeds, weights and loading configurations. All aircraft are designed to operate within a specific center of gravity range.

MAXIMUM FORWARD/MAXIMUM AFT C.G. LOCATIONS: Every aircraft has specified a forward most and rear most center of gravity location, along its longitudinal axis. These center of gravity location limits are given from a convenient reference (datum) on the aircraft.

DATUM: A convenient reference point along the longitudinal axis of an aircraft from which all horizontal measurements are taken.

WEIGHT: Actual individual weight of each item such as airframe, persons, fuel, baggage, cargo, etc., in pounds or kilograms.

ARM: Horizontal distance, along the longitudinal axis, measured between centroids of an object in the aircraft and the datum line.

MOMENT: Obtained by multiplying the weight of an item by its arm.

INSTALLED EQUIPMENT: All optional accessories and equipment permanently installed on an airframe or engine at the time of weighing. These items must be listed in the "List of Installed Equipment". Additions and deletions must be noted in the list each time they are made and new weight and balance calculations performed to determine the magnitude and effect of weight change. Ballast, if permanently installed, must also be listed.

BALLAST: A specific amount of weight attached in a specific location, which can be temporarily or permanently installed in an aircraft, to help bring its center of gravity within the required limits. If temporary ballast must be used for certain operations, the exact amount and its location must be placarded on the instrument panel within clear view of the pilot. The use of ballast increases empty weight and reduces useful load.

LOADING CHART: Used to calculate the actual center of gravity location of a ready to fly aircraft. Care must be taken not to exceed the maximum/minimum weight and balance limits stipulated for the aircraft. These limits are determined by structural, stability and control considerations of a particular design.

Sometimes it is necessary to adjust or reduce fuel, cargo, or passenger weights to remain at or below maximum allowable gross weight. A temporary or permanent ballast is sometimes necessary to bring the C.G. within specified limits. However, the maximum allowable gross weight should not be exceeded under any circumstances.

